STATE OF ISRAEL
Ministry of Transport
Civil Aviation Authority (CAA1)

TYPE CERTIFICATE for AIRCRAFT No. IA298

BRM Aero, s.r.o.
Letecka 255 686 04
Kunovice Czech Republic

This certificate issued to

certifies that the Type Design for the following product with the operating limitations and conditions herein as specified in the Air Navigation Regulations (Procedures for Documentation of Aircraft and Aircraft Parts) 1977, and the Type Certificate Data sheet, meets the airworthiness requirements of

BCAR SECTION S

Make
BRM Aero, s.r.o.

Model
BRISTELL RG

Note: See CAAI IA298 TCDS

This Certificate and the Type Certificate Data Sheet which is a part thereof shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Director General of the Civil Aviation Authority.

Date of application: December 17, 2016
Date of issuance: December 06, 2018

By the direction of the Director General, Civil Aviation Authority

Signed by: Benny Davidor
Title: Director, Airworthiness Division
### TRANSFER ENDORSEMENT

Immediately after transfer the person making the transfer shall notify the Civil Aviation Authority of Israel (CAA) of the name and address of the person to whom the certificate was transferred.

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This Data Sheet which is part of Type Certificate No. IA298 prescribes and conditions and limitations under which the product(s) for which the Type Certificate was granted meet(s) the standards of airworthiness required by the Israel Air Navigation Regulations.

**Type Certificate Holder:**
BRM Aero, s.r.o.
Letecká 255
686 04 Kunovice
Czech Republic

**I. GENERAL**

1. **Type designation:** BRISTELL RG
2. **Category:** Ultralight
3. **Submission date:** 17.12.2016
4. **Date of approval:** 06.12.2018

**II. REGULATORY BASIS**

1. **Airworthiness requirements:** CAP 482 (BCAR S) British Civil Airworthiness Requirements, Section S - Small Light Aeroplanes, Issue 6, 31 May 2013, plus MTOW increase to 600 kg in accordance with ASTM F2245-16.
2. **Special conditions:** No
3. **Exceptions:** No
III. TECHNICAL DATA, PERFORMANCE AND OPERATING LIMITATIONS


2. Description: The BRISTELL RG aeroplane is two-seat, cantilever low-wing monoplane of metal primary construction, with two side-by-side seats. The undercarriage is retractable tricycle with steerable nose wheel.

The aeroplane is fitted with 4-cylinder, 4-stroke Rotax 912 ULS (100 hp) engine and DUC Flashblack, 3-blade, composite propeller.


4. Basic technical data: Wing:
   Wing span............................................. 9.13 m
   Wing area.......................................... 11.5 m²
   Mean Aerodynamic Chord (MAC)............. 1.35 m
   Wing loading for MTOW=600 kg ........... 52.17 kg/m²
   Used wing airfoil............................. MS 316/314
   Aspect ratio...................................... 6.29
   Aileron span................................. 1.145 m
   Aileron area.................................. 0.29 m²
   Aileron deflections  up...................... +24 °
                        Down ......... -16 °
   Flap span................................. 2.3 m
   One flap area ...................... 0.828 m²
   Flap deflections............Take-off ...... 10 °
                        Landing .... 20,30 °
Fuselage:
   Length................................. 6.45 m
   Height ..................................... 2.28 m
   Width................................. 1.3 m
Horizontal Tail Unit (HTU):
   HTU span................................. 2.9 m
   HTU area.............................. 2.28 m²
   Elevator deflections.........Up .......... 90 °
                        Down.... 15 °
Vertical Tail Unit (VTU):
VTU height................................................. 1.08 m
VTU area.................................................. 0.93 m²
Rudder deflections.............................. +/- 30 °

Landing gear:
Main wheels track ......................... 1.88 m
Wheel base.............................................. 1.43 m
Wheel sizes:
Main wheels size............................... 355 mm
.................................................. 4.00-6 (Beringer)
Nose wheel size................................. 300 mm
.................................................. 4.00-4 (Beringer)

Hydropneumatic shock absorbers:
Main gear ...................... MARC P/N MI3213PB04
Nose gear .................. MARC P/N MI3220PA10

Tyre pressure................................. 180 + 20 kPa
Brakes........................................ Beringer dual brakes

5. Weights:
Maximum take-off weight (MTOW).......600 kg
Empty weight (standard equipment).....405 kg
Maximum empty weight................... 426 kg
(MTOW – 172 kg crew – 12 kg fuel for 1 hour flight at MCP)
Maximum usefull load....................... 195 kg
Minimum crew weight...................... 55 kg

Maximum baggage weight:
Baggage compartment „A“ behind seats ... 15 kg
Baggage compartment „B“ behind „A“ ... 1 kg
Max. baggage in wing lockers .......... 20 kg each

Fuel tanks (volume 2 x 60 l) .......... 120 l

6. Speed and performances:
Performance specifications under ISA, at MTOW 600 kg:
All speeds below are CAS if not stated otherwise:
Stall speed with deployed flaps \( V_{SO} \) ...... 39 mph
.................................................. 63 km/h
Stall speed with retracted flaps \( V_{S1} \) ...... 50 mph
.................................................. 81 km/h
Max. speed with extended flaps \( V_{FE} \) ...... 86 mph
.................................................. 139 km/h
Max. land gear operating speed \( V_{LO} \) ...... 86 mph
.................................................. 139 km/h
Design maneuvering speed $V_A$ ........... 112 mph
.............................................................................. 180 km/h
Max. land gear extend speed $V_{LE}$ .......... 112 mph
.............................................................................. 180 km/h
Maximum horizontal speed $V_H$ .......... 147 mph
............................................................................... 237 km/h
Never exceeded speed $V_{NE}$ ............ 173 mph (170 IAS)
............................................................................... 278 km/h (280 IAS)
Max. speed to use Magnum 601 ........ 203 mph
............................................................................... 327 km/h
Max. rate of climb ........................................ 4.4 m/sec
............................................................................... 860 ft/min
Takeoff distance over 15m obstacle ... 540 m

7. Center of gravity: Operation limits:
Front CG limit ......................................... 25 % MAC
Aft CG limit ............................................. 32 % MAC
Mean aerodynamic chord (MAC) ........ 1.350 m
Datum: Wing leading edge between ribs No. 4 and 5,
2071 mm (81.52 in) from plane of symmetry.

8. Load factors: Limit load factors:
Maximum positive .................................... +4 g
Maximum negative ................................... -2 g

9. Engine: Type/model: ROTAX 912 ULS
Producer: BRP-Powertrain GmbH & Co KG, Gunskirchen,
Austria
Take-off performance ...................... 73.5 kW at 5800 rpm
Max. continuous performance ........ 69 kW at 5500 rpm

10. Propeller and its limitations:
Type: DUC Flashblack
Producer: DUC Hélices Propellers, France
Description: in-flight adjustable, composite, 3 blade
Diameter: 1730 mm

11. Fuel: Obey the latest edition of Service Instruction SI-912-016, for
the selection of the correct fuel.
- Min. RON 95 (min. AKI4 91)
- EN 228 super
- EN 228 super plus
- AVGAS 100 LL

12. Oil: Oil type: At the selection of suitable lubricants refer to the
additional information in the Service Information SI-912-016,
latest edition

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Oil specification: Use only oil with RON 424 classification
Oil viscosity: Use of multi-grade oils is recommended

13. Coolant:

Obey the latest edition of Service Instruction SI-912-016, for the selection of the correct coolant.

Conventional e.g. BASF Glycantine anticorrosion, 50*:50 mixture of concentrate with water

* coolant component can be increased up to max. 65 %.
IV. DATA FOR OPERATION AND MAINTENANCE

1. Flight Manual: BRISTELL RG Aircraft Operating Instructions, Document No.: RG-AOI-2-10-1-IL, latest by CAAI approved revision


4. Propeller Manual: Instruction manual, Variable pitch propellers in flight, FLASHBLACK, revision supplied with the propeller


6. Instruments Equipment: Manuals and other documents supplied with installed instruments and equipment

V. SUPPLEMENTS:

None

Notes:
1. Aircraft must be fitted with labels and stickers in Hebrew and English language.
2. Fire extinguisher, First aid kit, and Emergency hammer with cutter must be installed in each aircraft

........END.......
BRISTELL RG Three view drawing